



Fatigue Crack Initiation in AA3004 At Elevated Temperature Using Rotating and Bending Fatigue Testing Machine

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Abstract

The numerous failures in engineering applications occur due to fatigue, therefore, design against fatigue failure is crucial and particularly fatigue failure of engineering applications having notches or holes because almost all applications have such type of discontinuities. Once the crack initiates it is very difficult to stop it, therefore, crack initiation in different materials is vital to be designed. This work is also related to the design of crack initiation in AA 3004 aluminum alloy at elevated temperature by means rotating and bending fatigue testing machine. A favorable working temperature environment has been set by means of the 1000W electric rod. The experimental results have been plotted in the form of SN curve and a comparison has been made between crack initiation at several mechanical loadings with and without thermal loading. The mathematical model for experimental data is also reported.

Keywords: Fatigue; Initiation; Aluminium alloy

Introduction

The alloy used in this research is an emerging material for recent applications [1]. Initiation and then propagation of crack causes the fatigue failure [2]. Crack propagation is led by the edge which is a source of stress concentration and it continues till failure [3]. Notches and holes have been observed in daily engineering applications. Crack initiation starts in such type of components and structures because they experience cyclic loading [4]. The existence of notches are unavoidable in materials and structures and a detailed knowledge for fatigue failure of the notched component under mechanical and thermal loading is significant [5]. Generally, there exists the thermal environment in materials and structures, therefore, a detailed knowledge is obligatory about what happens to fatigue crack under mechanical as well as thermal loading [6]. This study reveals the experimental investigation of fatigue crack initiation in the AA3004. The thermal loading is applied with a 1000W electric rod and the temperature range has been measured using a Multi-meter with a built-in temperature sensor.

Experimental Procedure

The specimen dimensions have been adjusted as per fatigue testing machine dimensions [7]. The specimen was prepared on a lathe machine. To ensure the avoidance of residual stresses the specimen was properly cooled down and specific tool speed was maintained. The PQ-6 fatigue testing machine equipped with 0.75kW motor and 3000r/min rotation speed which examines the fatigue strength of materials while rotating and bending. The repeatedly alternating bending force from hanging loads on supporting pads creates bending stress. Such type of experimentation was performed earlier on aluminum alloy [8]. Mechanical loading (ML) is applied ranging 160N–250N with a difference of 10N. Afterward, the specimen has been subjected to the thermal loading during single step mechanical loading. Upper and lower notch lie in the range of 50 °C–100 °C. Cross section temperature ranges 50 °C–60 °C after failure. During the second phase experiments, the varying load has been applied to the specimen.

Results & Discussion

The Eq.1 has been used to convert the applied load into stress,

$$\sigma = 509.3 \frac{Q}{d^3}, \quad (1)$$

As all the tested specimens have notches, Stress at notch is calculated by applying stress concentration factor Kt [9]

Nomenclature

- Ni = No. of Cycles to initiate a fatigue crack
- Np= No. of Cycles for a crack to propagate to final fracture
- S2 = Low Load
- S1 = High Load
- Nf2= Total life at low load
- Nf1= Total life at high load
- N2 = Number of cycles for S2 in two-step loading
- N1 = Number of cycles for S1 in two-step loading

After the effective start of crack initiation, it becomes difficult to hinder the crack growth. As the low load is applied after more than half of total fatigue life, crack was propagating in the duration of switching the load from low to high. Initially, low stress is applied in order to avoid strain hardening. If the sequence of applying stresses is changed then as a result whole initiation phase terminates in the same high level along with the starting of the next phase, complying the aim of cumulative damage accumulation. Initiation phase life is negligible, therefore, the number of cycles needed to end up the initiation phase are found from the results of experiments, in which remaining fraction of life after stress altering point “(1-x) Ni2” is compared with remaining cycles at high-stress level “yNi1”. Thus, the propagation life at low-stress level is compared with “1-y”. If the

value of ‘x’ is like, the initiation period at higher load is increased [10] then it can be concluded as;

$$xN_{f2} = N_{i2} + zN_{p2}, \quad (2)$$

$$x = \frac{N_2}{N_{f2}}, \quad (3)$$

$$y = \frac{N_1}{N_{f1}}, \quad (4)$$

$$yN_{f1} = yN_{p1}, as N_{i1} = 0, \quad (5)$$

$$z = 1 - y, \quad (6)$$

It is notified that the higher load conditions resulted in the increased crack speed and crack length at termination of initiation phase.

Hence,

$$xN_{f2} = N_{i2} + (1-y)(N_{f2} - N_{i2}), \quad (7)$$

Or

$$N_{i2} = \frac{N_{f2}(x+y-1)}{y}, \quad (8)$$

And the initiation phase life of the total life of all crack phases up to fracture [10]

$$\% \text{ of total life} = \frac{N_{i2}}{N_{f2}} \times 100 \quad (9)$$

Table 1 and Figure 1 show the data obtained as a result of applying only mechanical loading (ML) whereas Table 2 and Figure 2 show the data obtained as a result of applying both mechanical and thermal loadings (TL).

Table 1: Crack initiation life with only ML.

Exp. No.	S1	S2	N1	N2	Nf1	Nf2	x	y	x+y-1	Ni2	%Initiation
	[N]	[N]	[rpm]	[rpm]	[rpm]	[rpm]					
1	160	220	168695	504512	321178	557524	0,9	0,53	0,43	456594,58	81,9
2	170	220	251344	368424	321178	484141	0,76	0,78	0,54	336272,92	69,46
3	180	220	242496	403107	321178	487138	0,83	0,76	0,58	375841,7	77,15
4	190	220	239152	319397	321178	462030	0,69	0,74	0,44	270475,75	58,54
5	200	220	229641	344408	321178	447768	0,77	0,71	0,48	303207,76	67,72
6	210	220	159358	290713	321178	364263	0,8	0,5	0,29	216026,69	59,31
7	220	250	189361	222405	251981	321114	0,69	0,75	0,44	189762,81	59,1

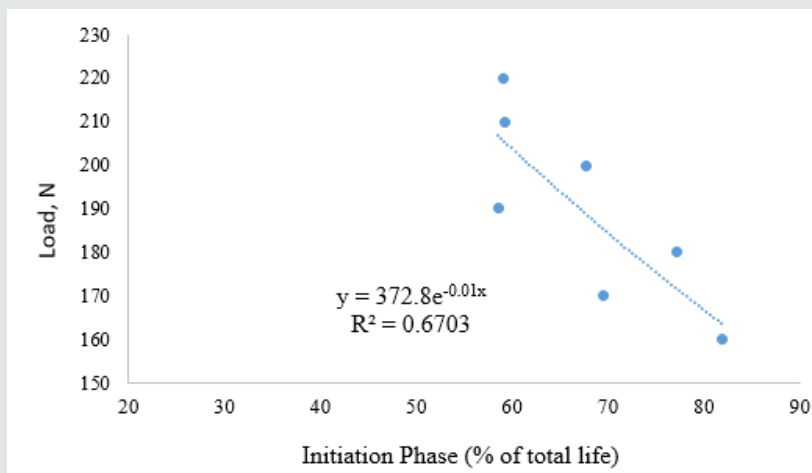


Figure 1: Load vs crack initiation life with only ML.

Table 2: Crack initiation life with both ML and TL.

Exp. No.	S1	S2	N1	N2	Nf1	Nf2	x	y	x+y-1	Ni2	% Initiation
	[N]	[N]	[rpm]	[rpm]	[rpm]	[rpm]					
1	160	220	101244	218895	108343	344769	0,63	0,93	0,57	210069	60,93
2	170	220	109485	171306	108343	291401	0,59	1,01	0,6	172558,67	59,22
3	180	220	92896	142568	108343	222960	0,64	0,86	0,5	129200,2	57,95
4	190	220	80567	124321	108343	206417	0,6	0,74	0,35	96017,87	46,52
5	200	220	75941	147827	108343	234903	0,63	0,7	0,33	110673,99	47,11
6	210	220	80993	87308	108343	151568	0,58	0,75	0,32	65608,46	43,29
7	220	250	29985	54593	39172	108279	0,5	0,77	0,27	38144,33	35,23

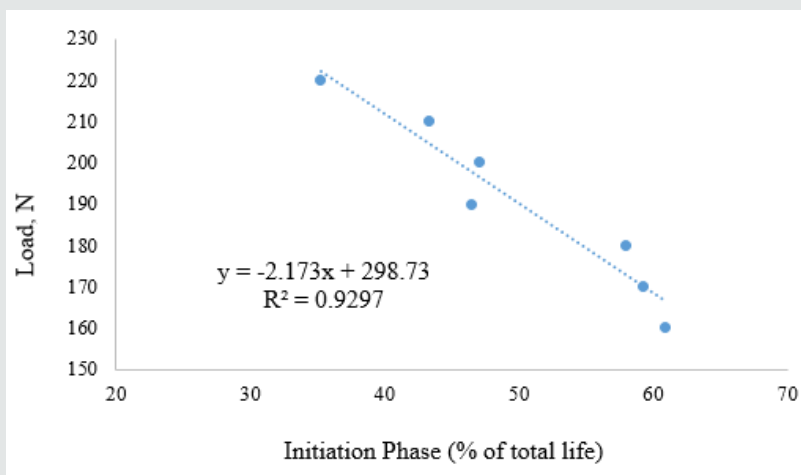


Figure 2: Load vs crack initiation life with ML and TL.

Conclusion

a. The initiation phase in the form %age of fatigue life of AA3004 aluminum alloy has been calculated which is clearly mentioned in graphical results.

b. The exponential behavior is obtained for the initiation of fatigue crack during only ML.

c. The linear behavior is obtained for the initiation of fatigue during the case of both ML and TL.

d. A mathematical model for the experimental data is represented in the form of equation mentioned in graphical results.

e. The number of cycles for crack initiation decrease by the addition of thermal loading.

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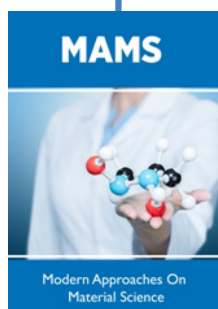
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